

computational effort required for estimation. Hence inaccuracies developed due to linearization are avoided.

2.2.4 Particle Filter

During the 1990s, the particle filter (PF) was well known in the field of recursive nonlinear state estimation, and has been widely applied in many fields (see e.g. (Gordon and al. [31], Bolviken and al. [32], Doucet and al. [33], Benhmida and al. [34]). The PF solves the Bayesian recursive relations by using Sequential Monte Carlo (SMC) methods. These methods allow for a complete representation of the posteriori probability density function of the states, so that any statistical estimates, such as the Minimum Mean Squared Error estimate (MMSE) and the Maximum a Posteriori Probabilities (MAP) can easily be computed. In year 2000, Kadirkamanathan and al. [35] introduced Sequential Monte Carlo methods in the field of fault detection and isolation (FDI). Because the PF is able to handle any functional nonlinearity as well as system or measurement noise of any probability distribution, it has attracted attention in the nonlinear non-Gaussian state estimation field.

2.2 Fault Detection and Isolation Techniques

2.2.1 FDI Using Multiple Model Based Adaptive Estimation (MMAE) Schemes

One approach to detect and isolate actuator or sensor faults is the multiple model adaptive estimation (MMAE) method [9]. It uses a bank of Kalman filters (KF) running in parallel, each of which is matching a particular fault status of the system. The next step is the use of a hypothesis testing algorithm by which the residuals from each KF is assigned a conditional probability for each fault scenario. It is evident that the computational load is quite intense. Thus the online use of this method was impractical for quite a long time. However, as time passed, more powerful processors were available making this method regain appeal in many applications. Several papers demonstrated how the MMAE method can be used in the context of fault detection and isolation for aircraft [10], [11], [12] and underwater vehicles [13]. The MMAE method will have reliable practical application as long as the expected faults can be hypothesized by a reasonable number of Kalman filters.

However, the major disadvantage is associated with the fact that the number of addressable faults is limited due to the computational load required for each filter.

2.2.2 FDI Using Extended Multiple Model Based Adaptive Estimation (EMMAE) Schemes

In order to make the MMAE method acceptable for any flight conditions and also capable of isolating lock-in-place or floating actuator faults, the MMAE algorithm is combined with Extended Kalman filters (EKF) which are capable of carrying out nonlinear estimation of some (unknown) fault parameter like the deflection of a faulty control surface (or actuator). The resulting method is called "Extended Multiple Model Adaptive Estimation" (EMMAE), [29], [30]. The EMMAE enables online estimation of the deflection of a faulty actuator to cope with lock-in-place or floating actuator

fault scenarios and drastically reduces the number of filters needed.

3. Fault Tolerant Control System

A fault-tolerant control system is capable of controlling the system with satisfactory performance even if one or several faults, or more critically, one or several failures occur in this system. Fault-tolerant control systems may be classified into two main categories: passive fault tolerant controllers and active fault-tolerant controllers.

3.1 Passive Fault Tolerant Controllers

In a passive fault-tolerant controller, when the plant parameters deviate from their true values or that of the actuators from their expected position, it may be efficiently compensated by a fixed robust feedback controller. However, if these deviations become excessively large and exceed the robustness properties, some actions are needed to be taken. Also, if deviations occur at the sensor side, inevitable deviations from the reference command signals will happen. Therefore, an active fault-tolerant control architecture is needed in order to achieve extended fault-tolerance capability and is thus preferred over passive controllers.

3.2 Active Fault Tolerant Controllers

An active fault-tolerant controller usually contains two separate modules. One module is basically a fault detection and isolation (FDI) system that monitors the health of the aircraft. The FDI system then informs the second module which is a supervision module, regarding the seriousness of the fault/failure or damage. Based thereon, the supervision module may decide to reconfigure the flight controllers, the guidance system, and the navigation system.

There are two families of FDI systems, namely passive FDI and active FDI systems. Passive FDI systems "wait" until something starts to clearly go wrong in the system [2], whereas the active FDI systems will artificially excite the aircraft, either by flying health check maneuvers [3], [4] or by injecting test signals [5], [6] in the actuator commands and then assessing the individual health status of actuators and sensors.

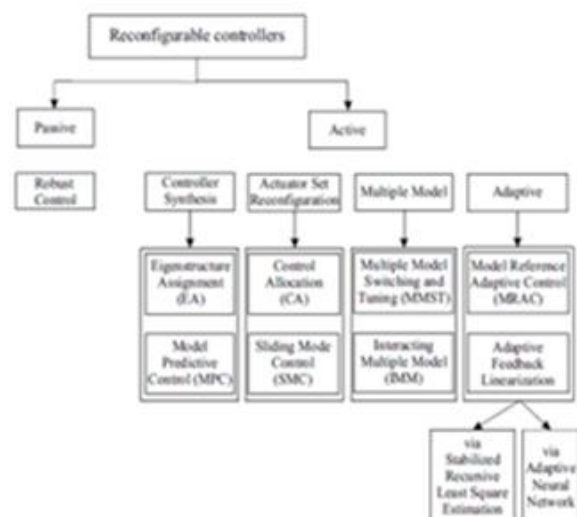


Figure 1: Fault Tolerant Control Schemes

4. Different Approaches for Fault Tolerant Control

The figure above, summarizes the most common techniques used in reconfigurable flight control systems. The mentioned techniques are explained below.

4.1 Multiple Model Techniques

4.1.1 Multiple Model Switching and Tuning (MMST)

In the MMST technique, the dynamics of each fault scenario are described by a dedicated model. Each model is also paired with its respective controller. When a fault occurs, the control system is reconfigured by choosing the model/controller pair that is the most appropriate at each time step. The MMST technique has the advantages of being fast and usually stable if the actually occurring failures match the predefined fault scenarios. However, severe limitations of the method arises in practice as soon as an unmodeled failure is encountered or if multiple or structural failures occur. Moreover, the number of individual pairs of model/controller to be designed may become excessively large if the system is to successfully operated over a wide range of failure scenarios [8], [7].

4.1.2 Multiple Model Adaptive Estimation (MMAE)

Another approach to detect and isolate actuator or sensor faults is the multiple model adaptive estimation (MMAE) method [9]. This method belongs to the family of Interacting Multiple Models (IMM). It is based on a bank of Kalman filters (KF) running in parallel, each of which is matching a particular fault status of the system. A hypothesis testing algorithm then uses the residuals from each KF to assign a conditional probability to each fault hypothesis. As one may expect, the computational load is quite intense. Therefore, the online use of this method was impractical for a long time. However, with the more powerful processors now available this method has regained appeal in many applications.

Several papers have demonstrated how the MMAE method can be used in the context of fault detection and isolation and control reconfiguration for aircraft [10], [11], [12] and underwater vehicles [13].

4.2 Control Allocation Techniques

Control allocation techniques can be briefly stated by the following general relation, the flight control system generates a virtual control command $CV = [CL \ CM \ CN]^T$ in terms of the desired roll, pitch, and yaw torques. This virtual command CV is passed to the control allocator which is provided for each actuator's position limits and effectiveness to produce any torque component of the CV vector. An algorithm is then computed online to optimally generate control signals for the actuators [14], [15], [16].

The major advantage of using a control allocation technique is that actuator failures can be compensated for without the need for modifying the flight control laws [17]. Moreover, actuator's constraints, such as deflection limits and motion

rates, can be taken into account by the control allocator (CA) when the virtual command CV is "distributed" over the actuators. Finally, the deflection of each actuator can be chosen by the CA to optimize some criteria, such as total drag, total deflections, or to prioritize some actuators. However, as explained in [7], control allocation techniques may have the following possible disadvantage: "the dynamics and limitations of the actuators after a failure are not taken into account in the control laws." This means that the controller will still attempt to achieve the original system performance even though the actuators are not capable of achieving it.

4.3 Model Reference Adaptive Control

Model reference adaptive control [18], [19] is a method which can be utilized when tolerance to damage or structural failures is required. This technique is also often used as a final stage of a complex control system combining several algorithms. The working principle of this type of control is to have the output of the plant under consideration to follow the output of a reference model. However, the Model Reference Adaptive Control (MRAC) technique has some limitations. Firstly, the adaptation laws require an estimation algorithm to track certain parameters of the system. It is therefore necessary that these system parameters evolve slowly enough in order that the estimation routine can track them properly. Faults or failures, however, may cause abrupt changes in the values of the system parameters. Secondly, during the transient phase in which the adaptive algorithm identifies the new faulty plant; it is not guaranteed that the controller can stabilize the system. Therefore, the model reference adaptive control technique is usually not used on its own but in combination with other algorithms in a more complicated fault-tolerant control architecture [20], [21].

4.4 Other Reconfigurable Control Methods

There are other methods to design a reconfigurable flight control system. For instance, the Eigen structure assignment (EA) is used to reconfigure the feedback control laws in [22] and [23]. In Model Predictive Control (MPC), the constraints on actuators or on any other state variables are systematically taken into account during the generation of the control signals [24]. Sliding Mode Control (SMC) has been investigated in [25] or more recently in [13], [26], and [27]. Other reconfigurable flight control systems use adaptive feedback linearization via artificial neural network (ANN) [28] or via online parameter identification methods [20].

5. Conclusion

In this review paper, various fault detection and fault tolerant control techniques associated with flight were discussed. The reviewed FDI techniques reveal that model based fault detection and estimation can be used successfully for determining the actuator, control surface or sensor faults in aircrafts. Similarly, the FTC techniques proved to be sophisticated when they are used as a combination of several algorithms to improve the fault associated scenario. These techniques, both FDI and FTC for aircrafts can be applied to space vehicles like launch vehicles, spacecraft and reusable

launch vehicles. This is mainly due to the fact that they follow the same dynamics of flight when they make reentry into the Earth's atmosphere.

References

- [1] R. Isermann. Fault-Diagnosis Systems, An Introduction from Fault Detection to Fault Tolerance. Springer-Verlag, Berlin Heidelberg, 2006.
- [2] G. Hearn, M. J. Grimble, and M. A. Johnson. Integrated Fault Monitoring and Reliable Control. In UKACC International Conference on CONTROL'98, pages 1175–1179, 1-4 September 1998.
- [3] M. Azam, K. Pattipati, J. Allanach, S. Poll, and A. Petterson-Hine. In-Flight Fault Detection and Isolation in Aircraft Flight Control Systems. In Proceedings of IEEE Aerospace Conference, paper 1429, 2005.
- [4] M. Elgersma, D. Enns, S. Shald, and P. Voulgaris. Parameter Identification for Systems with Redundant Actuators. In Proceedings of the AIAA Guidance, Navigation and Control Conference and Exhibit, Boston, 2004.
- [5] S. L. Campbell and R. Nikoukhah. Auxiliary Signal Design for Failure Detection. Princeton University Press, New Jersey, 2004.
- [6] G. Ducard and H. P. Geering. A Reconfigurable Flight Control System based on the EMMAE Method. In Proceedings of the IEEE American Control Conference, pages 5499–5504, Minneapolis, MN, June 2006.
- [7] C. N. Jones. Reconfigurable Flight Control, First Year Report. Available at <http://www.control.eng.cam.ac.uk/cnj22/docs/yearone.pdf>, September 2002.
- [8] J. D. Boskovic and R. K. Mehra. A Multiple Model-based Reconfigurable Flight Control System Design. In Proceedings of the 37th IEEE Conference on Decision and Control, pages 4503–4508, Tampa, FL, December 1998.
- [9] D. T. Magill. Optimal Adaptive Estimation of Sampled Stochastic Processes. IEEE Transactions on Automatic Control, 10(4):434–439, October 1965.
- [10] P. Eide and P. S. Maybeck. An MMAE Failure Detection System for the F-16. IEEE Transactions on Aerospace and Electronic Systems, 32(3):1125–1136, July 1996.
- [11] P. S. Maybeck. Multiple Model Adaptive Algorithms for Detecting and Compensating Sensor and Actuator/Surface Failures in Aircraft Flight Control Systems. International Journal of Robust and Nonlinear Control, 9(14):1051–1070, 1999.MA, August 1998.
- [12] P. S. Maybeck and R. D. Stevens. Reconfigurable Flight Control via Multiple Model Adaptive Control Methods. IEEE Transactions on Aerospace and Electronic Systems, 27(3):470–479, May 1991.
- [13] L. Ni. Fault-Tolerant Control of Unmanned Underwater Vehicles. PhD thesis, VA Tech. Univ., Blacksburg, VA, 2001.
- [14] M. Bodson. Evaluation of Optimization Methods for Control Allocation. AIAA Journal of Guidance, Control, and Dynamics, 25(4):703–711, 2002.
- [15] J. Buffington and D. Enns. Lyapunov Stability Analysis of Daisy Chain Control Allocation. AIAA Journal of Guidance, Control, and Dynamics, 19(6):1226–1230, 1996.
- [16] W. Durham. Constrained Control Allocation. AIAA Journal of Guidance, Control, and Dynamics, 16(4):717–725, 1993.
- [17] O. Harkegard. Backstepping and Control Allocation with Applications to Flight Control. PhD thesis, Linköping University, Sweden, 2003.
- [18] E. Shafai. Einführung in die Adaptive Regelung. Lecture Notes, IMRT, ETH Zurich, 2003.
- [19] G. Tao, S. Chen, X. Tang, and S. M. Joshi. Adaptive Control of Systems with Actuator Failures. Springer-Verlag, London Berlin Heidelberg, 2004.
- [20] M. Bodson. Reconfigurable Nonlinear Autopilot. AIAA Journal of Guidance, Control, and Dynamics, 26(5):719–727, 2003.
- [21] J. D. Schierman, D. G. Ward, J. R. Hull, N. Gandhi, M. W. Oppenheimer, and D. B. Doman. Integrated Adaptive Guidance and Control for Re-Entry Vehicles with Flight-Test Results. AIAA Journal of Guidance, Control, and Dynamics, 27(6):975–988, November-December 2004.
- [22] Y. Zhang and J. Jiang. Integrated Active Fault-Tolerant Control Using IMM Approach. IEEE Transactions on Aerospace and Electronic Systems, 37(4):1221–1235, October 2001.
- [23] Y. M. Zhang and J. Jiang. Active Fault Tolerant Control System against Partial Actuator Failures. IEEE Proceedings in Control Theory Application, 149(1):95–104, January 2001.
- [24] J. M. Maciejowski and C. N. Jones. MPC Fault-Tolerant Flight Control Case Study: Flight 1862. In Proceedings of the IFAC SAFEPROCESS Conference, Washington DC, June 2003.
- [25] Y. B. Shtessel and J. Buffington. Multiple Time Scale Flight Control using Reconfigurable Sliding Modes. AIAA Journal of Guidance, Control, and Dynamics, 22(6):873–883, November-December 1999.
- [26] H. Alwi and C. Edwards. Fault Tolerant Control of a Civil Aircraft Using a Sliding Mode Based Scheme. In Proceedings of IEEE Control and Decision Conference, and European Control Conference, pages 1011–1016, Seville, Spain, 2005.
- [27] M. L. Corradini, G. Orlando, and G. Parlangeli. A Fault Tolerant Sliding Mode Controller for Accommodating Actuator Failures. In Proceedings of IEEE Control and Decision Conference, and European Control Conference, pages 3091–3096, Seville, Spain, 2005.
- [28] A. J. Calise, S. Lee, and M. Sharma. Direct Adaptive Reconfigurable Control of a Tailless Fighter Aircraft. In Proceedings of the AIAA Guidance, Navigation, and Control Conference and Exhibit, Boston, MA, August 1998.
- [29] G. Ducard and H. P. Geering. A Reconfigurable Flight Control System based on the EMMAE Method. In Proceedings of the IEEE American Control Conference, pages 5499–5504, Minneapolis, MN, June 2006.
- [30] D. Rupp, G. Ducard, H. P. Geering, and E. Shafai. Extended Multiple Model Adaptive Estimation for the Detection of Sensor and Actuator Faults. In Proceedings of IEEE Control and Decision Conference, and European Control Conference, pages 3079–3084, Seville, Spain, 2005.
- [31] N. Gordon, D. J. Salmond and A. F. M. Smith, Novel approach to nonlinear/non-Gaussian Bayesian state estimation, IEE Proceedings-F, 140(2): 107- 113,1993.

- [32] E. Bolviken, P. J. Acklam, N. Christophersen, and J-M. Stordal, Monte Carlo filters for non-linear state estimation, *automatic*, 37(2): 177-183, 2001.
- [33] A. Doucet, N. Freitas, and N. Gordon, *Sequential Monte Carlo Methods in Practice* Springer-Verlag, New York, 2001.
- [34] F. Ben Hmida, F. Souibgui and A. Chaari, Estimation d'état des stochastiques non linéaires par l'approche bayésienne récursive, 8th conférence internationale STA07, 05-07 novembre à Mo-nastir-TUNISIE, 2007.
- [35] Kadiramanathan, V., P. Li, M.H. Jaward, and S.G. Fabri. Particle filtering-based fault detection in nonlinear stochastic systems. *International Journal of Systems Science* (33), 259265, 2002.

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